



Standard Test Method for In-Plane Shear Response of Polymer Matrix Composite Materials by Tensile Test of a $\pm 45^\circ$ Laminate¹

This standard is issued under the fixed designation D 3518/D 3518M; the number immediately following the designation indicates the year of original adoption or, in the case of revision, the year of last revision. A number in parentheses indicates the year of last reapproval. A superscript epsilon (ϵ) indicates an editorial change since the last revision or reapproval.

This standard has been approved for use by agencies of the Department of Defense.

1. Scope

1.1 This test method determines the in-plane shear response of polymer matrix composite materials reinforced by high-modulus fibers. The composite material form is limited to a continuous-fiber-reinforced composite $\pm 45^\circ$ laminate capable of being tension tested in the laminate χ direction.

1.2 *This standard does not purport to address all of the safety concerns, if any, associated with its use. It is the responsibility of the user of this standard to establish appropriate safety and health practices and determine the applicability of regulatory limitations prior to use.*

1.3 The values stated in either SI units or inch-pound units are to be regarded separately as standard. Within the text the inch-pound units are shown in brackets. The values stated in each system are not exact equivalents; therefore, each system must be used independently of the other. Combining values from the two systems may result in nonconformance with the standard.

2. Referenced Documents

2.1 ASTM Standards:

- D 883 Terminology Relating to Plastics²
- D 3039/D 3039M Test Method for Tensile Properties of Polymer Matrix Composite Materials³
- D 3878 Terminology for High-Modulus Reinforcing Fibers and Their Composites³
- D 5229/D 5229M Test Method for Moisture Absorption Properties and Equilibrium Conditioning of Polymer Matrix Composite Materials³
- E 6 Terminology Relating to Methods of Mechanical Testing⁴
- E 111 Test Method for Young's Modulus, Tangent Modulus, and Chord Modulus⁴
- E 177 Practice for Use of the Terms Precision and Bias in

ASTM Test Methods⁵

- E 456 Terminology Relating to Quality and Statistics⁵
- E 1309 Guide for Identification of Composite Materials in Computerized Material Property Databases³
- E 1313 Guide for Recommended Formats for Data Records Used In Computerization of Mechanical Test Data for Metals⁶
- E 1434 Guide for Development of Standard Data Records for Computerization of Mechanical Test Data for High-Modulus Fiber-Reinforced Composite Materials³
- E 1471 Guide for Identification of Fibers, Fillers, and Core Materials in Computerized Material Property Databases³

3. Terminology

3.1 *Definitions*—Terminology D 3878 defines terms relating to high-modulus fibers and their composites. Terminology D 883 defines terms relating to plastics. Terminology E 6 defines terms relating to mechanical testing. Terminology E 456 and Practice E 177 define terms relating to statistics. In the event of a conflict between terms, Terminology D 3878 shall have precedence over the other standards.

3.2 *Definitions of Terms Specific to This Standard:*

NOTE 1—If the term represents a physical quantity, its analytical dimensions are stated immediately following the term (or letter symbol) in fundamental dimension form, using the following ASTM standard symbology for fundamental dimensions, shown within square brackets: [M] for mass, [L] for length, [T] for time, [Θ] for thermodynamic temperature, and [nd] for nondimensional quantities. Use of these symbols is restricted to analytical dimensions when used with square brackets, as the symbols may have other definitions when used without the brackets.

3.2.1 $\pm 45^\circ$ laminate—in laminated composites, a balanced, symmetric lay-up composed only of $+45^\circ$ plies and -45° plies. (See also *ply orientation*.)

3.2.2 *balanced, adj*—in laminated composites, having, for every off-axis ply oriented at $+\theta$, another ply oriented at $-\theta$ that is of the same material system and form.

3.2.3 *lamina, n—pl. laminae, in laminated composites*, a single, thin, uniform layer that is the basic building block of a laminate. (Syn. *ply*).

¹ This test method is under the jurisdiction of ASTM Committee D30 on Composite Materials and is the direct responsibility of Subcommittee D30.04 on Lamina and Laminate Test Methods.

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² *Annual Book of ASTM Standards*, Vol 08.01.

³ *Annual Book of ASTM Standards*, Vol 15.03.

⁴ *Annual Book of ASTM Standards*, Vol 03.01

⁵ *Annual Book of ASTM Standards*, Vol 14.02.

⁶ *Annual Book of ASTM Standards*, Vol 14.01.

3.2.4 *material coordinate system, n*—in laminated composites, a 123 Cartesian coordinate system describing the principle material coordinate system for a laminated material, where the 1-axis is aligned with the ply principal axis, as illustrated in Fig. 1. (See also *ply orientation, ply principal axis, and principal material coordinate system.*)

3.2.5 *nominal value, n*—a value, existing in name only, assigned to a measurable property for the purpose of convenient designation. Tolerances may be applied to a nominal value to define an acceptable range for the property.

3.2.6 *off-axis, adj*—in laminated composites, having a ply orientation that is neither 0 nor 90°.

3.2.7 *ply, n*—in laminated composites, synonym for *lamina*.

3.2.8 *ply orientation, n, θ*—in laminated composites, the angle between a reference direction and the ply principal axis. The angle is expressed in degrees, greater than -90° but less than or equal to +90°, and is shown as a positive quantity when taken from the reference direction to the ply principal axis, following the right-hand rule.

3.2.8.1 *Discussion*—The reference direction is usually related to a primary load-carrying direction.

3.2.9 *ply principal axis, n*—in laminated composites, the coordinate axis in the plane of each lamina that defines the ply orientation. (See also *ply orientation and material coordinate system.*)

3.2.9.1 *Discussion*—The ply principal axis will, in general, be different for each ply of a laminate. The angle that this axis makes relative to a reference axis is given by the ply orientation. The convention is to align the ply principal axis with the direction of maximum stiffness (for example, the fiber direction of unidirectional tape or the warp direction of fabric reinforced material).

3.2.10 *principal material coordinate system, n*—a coordinate system having axes that are normal to planes of symmetry within the material. (See also *material coordinate system.*)

3.2.10.1 *Discussion*—Common usage, at least for Cartesian coordinate systems (for example, 123 or *xyz*), aligns the first axis of the principal material coordinate system with the direction of highest property value; for elastic properties, the axis of greatest elastic modulus is aligned with the 1 or *x* axes.

3.2.11 *symmetric, adj*—in laminated composites, when the constituents, material form, and orientation for the plies located on one side of the laminate midplane are the mirror image of the plies on the other side of the midplane.

3.2.12 *transition region, n*—a strain region of a stress-strain or strain-strain curve over which a significant change in the slope of the curve occurs within a small strain range.

3.2.12.1 *Discussion*—Many filamentary composite materials exhibit a nonlinear stress/strain response during loading, such as seen in plots of either longitudinal stress versus longitudinal strain or transverse strain versus longitudinal strain. In certain cases, the nonlinear response may be conveniently approximated by a bilinear fit. There are varying physical reasons for the existence of a transition region. Common examples include matrix cracking under tensile loading and ply delamination.

3.3 *Symbols:*

3.3.1 *A*—cross-sectional area of a coupon.

3.3.2 *CV*—coefficient of variation statistic of a sample population for a given property (in percent).

3.3.3 F_{12}° (*offset*)—the value of the τ_{12} shear stress at the intersection of the shear chord modulus of elasticity and the stress stress curve, when the modulus is offset along the shear strain axis from the origin by the reported strain offset value.

3.3.4 G_{12} —in-plane shear modulus of elasticity.

3.3.4.1 *Discussion*—Indices 1 and 2 indicate the fiber direction and transverse to the fiber direction in the plane of the ply, respectively, as illustrated in Fig. 2.

3.3.5 *n*—number of coupons per sample population.

3.3.6 *P*—load carried by test coupon.

3.3.7 P^m —the load carried by test coupon that is the lesser of the (1) maximum load before failure or (2) load at 5 % shear strain.

3.3.8 s_{n-1} —standard deviation statistic of a sample population for a given property.

3.3.9 χ_i —test result for an individual coupon from the sample population for a given property.

3.3.10 $\bar{\chi}$ —mean or average (estimate of mean) of a sample population for a given property.

3.3.11 ϵ —general symbol for strain, whether normal strain or shear strain.

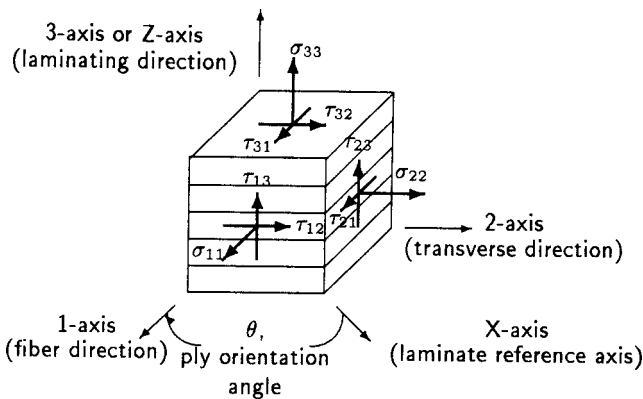
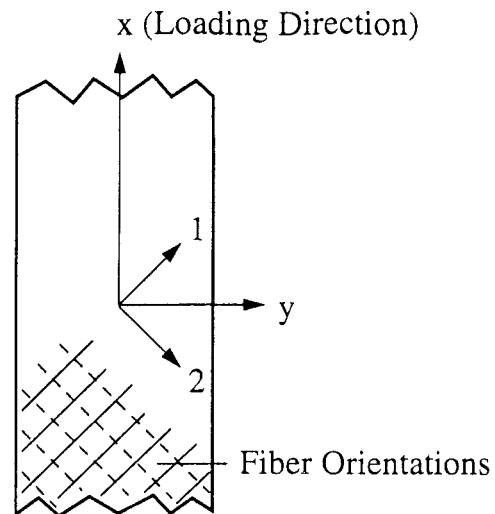


FIG. 1 Material Coordinate System



x and *y* represent the Specimen or Reference Axes, while 1 and 2 represent the Material or Local Axes

FIG. 2 Definition of Specimen and Material Axes

3.3.12 ϵ —indicated normal strain from strain transducer or extensometer.

3.3.13 τ_{12} —shear stress on the plane perpendicular to the 1-axis that acts parallel to the 2-axis.

3.3.14 τ_{12}^m —the calculated value of the τ_{12} shear stress taken at the lesser of (1) maximum shear stress before failure or (2) shear stress at 5 % shear strain.

3.3.15 γ_{12} —shear strain on the plane perpendicular to the 1-axis that acts parallel to the 2-axis.

3.3.16 γ_{12}^m —the value of the γ_{12} shear strain at the maximum shear stress before failure, or 5 %, whichever is less.

4. Summary of Test Method

4.1 A uniaxial tension test of a $\pm 45^\circ$ laminate is performed in accordance with Test Method D 3039, although with specific restrictions on stacking sequence and thickness. Use of this test for evaluation of in-plane shear response was originally proposed by Petit⁷ and was later improved by Rosen.⁸ Using expressions derived from laminated plate theory, the in-plane shear stress in the material coordinate system is directly calculated from the applied axial load, and the related shear stress is determined from longitudinal and transverse normal strain data obtained by transducers. This data is used to create an in-plane shear stress-shear strain curve.

5. Significance and Use

5.1 This test method is designed to produce in-plane shear property data for material specifications, research and development, quality assurance, and structural design and analysis. Factors that influence the shear response and should therefore be reported include the following: material, methods of material preparation and lay-up, specimen stacking sequence and overall thickness, specimen preparation, specimen conditioning, environment of testing, specimen alignment and gripping, speed of testing, time at temperature, void content, and volume percent reinforcement. Properties that may be derived from this test method include the following:

- 5.1.1 In-plane shear stress versus shear strain response,
- 5.1.2 In-plane shear chord modulus of elasticity,
- 5.1.3 Offset shear properties,
- 5.1.4 Maximum in-plane shear stress for a $\pm 45^\circ$ laminate, and
- 5.1.5 Maximum in-plane shear strain for a $\pm 45^\circ$ laminate.

6. Interferences

6.1 *Impurity of Stress Field*—The material in the gage section of this specimen is not in a state of pure in-plane shear stress, as an in-plane normal stress component is present throughout the gage section and a complex stress field is present close to the free edges of the specimen. Although this test method is believed to provide reliable initial material

response and can establish shear stress-shear strain response well into the nonlinear region, the calculated shear stress values at failure do not represent true material strength values and should only be used with caution. Despite attempts to minimize these effects, the shear stress at failure obtained from this test method, even for otherwise identical materials that differ only in cured ply thickness or fabric areal weight, may have differing failure modes and may not be able to be statistically pooled. The technical basis for the further discussion below is taken from the paper by Kellas et al.⁹

6.1.1 *Effects of In-Plane Normal Stress Field*—Of particular concern is the in-plane stress component normal to the fiber direction. This component of stress is present in all plies and throughout the gage section of the specimen. The effect of this stress on a given ply is minimized by the fiber reinforcement of the neighboring plies. Since the ply constraint is reduced with increasing ply thickness, the thickness of the individual plies is an important parameter that influences both the shear stress-shear strain response and the ultimate failure load of this specimen.¹⁰ Moreover, the surface plies of a given specimen, being constrained by only one neighboring ply (as opposed to interior plies, which are constrained by a ply on each side), represent the weakest link in a $\pm 45^\circ$ specimen. During the tensile loading of this test coupon, the first ply failures consist primarily of normal stress (or mixed mode) failures, rather than pure shear failures. Because of this, the actual material shear strength cannot be obtained from this test. Except for the case of materials capable of sustaining large axial test coupon strains (greater than about 3.0 %), the shear stress at failure is believed to underestimate the actual material shear strength.

6.1.2 *Total Thickness Effects*—As a result of the failure processes discussed above, the shear stress-shear strain response at higher strain levels depends upon the total number of plies. As the total number of plies in the specimen configuration is increased, the relative contribution of the two weak surface plies to the total load-carrying capacity is decreased. After the surface plies of the laminate fail, their portion of the load is redistributed to the remainder of the intact plies. The higher the total number of plies, the greater the chance that the remaining plies will be able to carry the load without immediate ultimate failure of the coupon. However, with each successive ply matrix failure the number of remaining intact plies diminishes, to the point where the applied load can no longer be carried. Because of this process, higher ply count specimens tend to achieve higher failure loads. To minimize these effects, this test method requires the use of a homogeneous stacking sequence and requires a fixed number of plies, for which the only repeating plies are the two required for symmetry on opposite sides of the laminate mid plane.

6.1.3 *Effects of Large Deformation*—Note that extreme fiber scissoring can occur in this specimen for the cases of ductile

⁷ Petit, D. H., "A Simplified Method of Determining the In-plane Shear Stress/Strain Response of Unidirectional Composites," *Composite Materials: Testing and Design, ASTM STP 460*, American Society for Testing and Materials, 1969, pp. 83–93.

⁸ Rosen, B. W., "A Simple Procedure for Experimental Determination of the Longitudinal Shear Modulus of Unidirectional Composites," *Journal of Composite Materials*, October 1972, pp. 552–554.

⁹ Kellas, S., Morton, J., and Jackson, K. E., "Damage and Failure Mechanisms in Scaled Angled-Ply Laminates," *Fourth Composites Symposium on Fatigue and Fracture, ASTM STP 1156*, W. Stinchcomb and Ashbaugh, N. E., Eds., American Society for Testing and Materials, 1993, pp. 257–280.

¹⁰ Repeating plies (adjacent plies at the same ply orientation) have an effect similar to thick plies, therefore, this test method prohibits constructions with repeating plies.

matrices, weak fiber/matrix interfaces, thick specimens with a large number of repeated plies, or a combination of the above. Kellas et al suggest that a general rule of thumb for this specimen is that a fiber rotation of 1° takes place for every 2 % of axial strain (or every 3.5 % shear strain for commonly tested materials). Such fiber scissoring, if left unbounded, would lead to an unacceptable violation of the assumption in this test method of a nominal ±45° laminate. This is the principal rationale for terminating this test at a large strain level, even if load is still increasing on the specimen. This test method terminates data reporting at 5 % calculated shear strain; this limits fiber scissoring to about 1.5°, is approximately the limit of foil strain gage technology (if used), and is also well beyond the strain levels required for common engineering practice. Further details of the effects of stacking sequence, specimen geometry, and, in particular, specimen and ply thickness, are presented in the reference by Kellas et al.

6.1.4 Effects of Edge Stresses—Even though interlaminar stresses reach a maximum value near the free edges of this laminate, the effect of interlaminar stresses on the failure process of ±45° laminates is insignificant when compared to the effect of the normal stress component transverse to the fiber direction in the plane of the specimen. Therefore, the effect of specimen width is much less important than stacking sequence and specimen thickness effects.

6.1.5 Effect of Axial Stress Nonuniformity—Both the shear stress and the shear modulus calculations depend upon the uniformity of the applied axial stress. Since the average applied load is used to calculate the shear stress this will not necessarily correspond to the stress in the vicinity of the measured shear strain, unless the axial stress is uniform throughout the volume of the stressed material. Therefore, the greater the degree of material inhomogeneity, such as with coarsely woven fabrics or materials with significant resin-rich regions, the greater the potential for inaccuracies in the measured response.

6.2 Other—Additional sources of potential data scatter in testing of composite materials are described in Test Method D 3039.

7. Apparatus

7.1 Apparatus shall be in accordance with Test Method D 3039. However, this test method requires that load-normal strain data be measured in both the longitudinal and transverse directions of the coupon.

8. Sampling and Test Specimens

8.1 Sampling—Sampling shall be in accordance with Test Method D 3039.

8.2 Geometry—The coupon geometry shall be in accordance with Test Method D 3039, as modified by the following:

8.2.1 The stacking sequence shall be [45/−45]*n*s, where 4 ≤ *n* ≤ 6 for unidirectional tape (16, 20, or 24 plies) and 2 ≤ *n* ≤ 4 for woven fabric (8, 12, or 16 plies). The recommended coupon width is 25 mm [1.0 in.], and the recommended coupon length range is 200 to 300 mm [8 to 12 in.], inclusive.

NOTE 2—Tabs, which are optional for the Test Method D 3039/D 3039M test coupon, are normally not required for successful conduct of this Practice.

8.3 Specimen Preparation—Specimen preparation shall be

in accordance with Test Method D 3039/D 3039M.

9. Calibration

9.1 Calibration shall be in accordance with Test Method D 3039/D 3039M.

10. Conditioning

10.1 Conditioning shall be in accordance with Test Method D 3039/D 3039M.

11. Procedure

11.1 Perform a tension test on the ±45° laminate coupon in accordance with Test Method D 3039/D 3039M, with normal strain instrumentation in both longitudinal and transverse directions and continuous or nearly continuous load-normal strain data recording. If ultimate failure does not occur within 5 % shear strain, the data shall be truncated to the 5 % shear strain mark (see 6.1.3 for the explanation). When the data is truncated, for the purpose of calculation and reporting, this 5 % shear strain point shall be considered the maximum shear stress. Any truncation of data shall be noted in the report.

12. Calculation

12.1 Maximum Shear Stress/Shear Stress— Calculate the maximum in-plane shear stress for the ±45° laminate using Eq 1 and report the results to three significant figures. If the shear modulus is to be calculated, determine the shear stress at each required data point using Eq 2.

$$\tau_{12}^m = \frac{P^m}{2A} \quad (1)$$

$$\tau_{12i} = \frac{P_i}{2A} \quad (2)$$

where:

τ_{12}^m = maximum in-plane shear stress, MPa [psi];

P^m = maximum load at or below 5 % shear strain, N [lbf];

τ_{12i} = shear stress at *i*-th data point, MPa [psi];

P_i = load at *i*-th data point, N [lbf]; and

A = cross-sectional area in accordance with Test Method D 3039/D 3039M, mm² [in.²].

12.2 Shear Strain/Maximum Shear Strain— If shear modulus or maximum shear strain is to be calculated, determine the shear strain at each required data point using Eq 3. The maximum shear strain is determined from Eq 4. Report the results to three significant figures.

$$\gamma_{12i} = \epsilon_{x_i} - \epsilon_{y_i} \quad (3)$$

$$\gamma_{12}^m = \min \left\{ \begin{matrix} 5\% \\ \gamma_{12i} \text{ at maximum shear stress} \end{matrix} \right. \quad (4)$$

where:

γ_{12i} = shear strain at *i*-th data point, με;

ϵ_{x_i} = longitudinal normal strain at *i*-th data point, με; and

ϵ_{y_i} = lateral normal strain at *i*-th data point, με; and

γ_{12}^m = maximum shear strain, με.

12.3 Shear Modulus of Elasticity:

12.3.1 Chord Shear Modulus of Elasticity—Calculate the chord shear modulus of elasticity using Eq 5, applied over a 4000 ± 200 με shear strain range, starting with the lower strain

point in the range of 1500 to 2500 $\mu\epsilon$, inclusive. Report the chord shear modulus of elasticity to three significant figures. Also report the shear strain range used in the calculation. A graphical example of chord shear modulus is shown in Fig. 3.

NOTE 3—The shear strain range of 2000 to 6000 $\mu\epsilon$ for shear modulus determination was selected, based on the shear response of a $\pm 45^\circ$ tensile coupon with a Poisson ratio near 1.0, to correspond approximately to the normal strain range of 1000 to 3000 $\mu\epsilon$ used to report the tensile chord modulus of elasticity in Test Method D 3039/D 3039M.

12.3.1.1 A different strain range must be used for materials that fail or exhibit a transition region (a significant change in the slope of the stress-strain curve) before 6000 $\mu\epsilon$. In such cases, the upper strain range value for the sample population shall be determined after testing; defined as 90 % of the average value of the upper limit of the essentially linear region, rounded downward to the nearest 500 $\mu\epsilon$. Any presence of a transition region shall be reported, along with the strain range used.

$$G_{12}^{\text{chord}} = \frac{\Delta\tau_{12}}{\Delta\gamma_{12}} \quad (5)$$

where:

- G^{chord} = shear chord modulus of elasticity, GPa [psi];
- $\Delta\tau_{12}$ = difference in applied shear stress between the two shear strain points, MPa [psi]; and
- $\Delta\gamma_{12}$ = difference between the two shear strain points (nominally 0.004).

12.3.2 *Shear Modulus of Elasticity (Other Definitions)*—Other definitions of elastic modulus may be evaluated and reported at the user’s discretion. If such data is generated and reported, report also the definition used, the shear strain range used, and the results to three significant figures. Test Method E 111 provides additional guidance in the determination of modulus of elasticity.

NOTE 4—An example of another modulus definition is the secondary

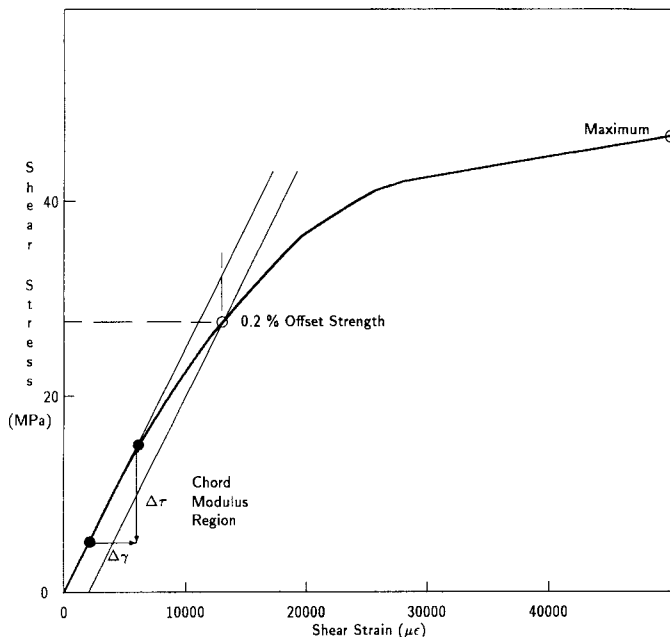


FIG. 3 Illustration of Modulus and Offset Strength Determination

chord modulus of elasticity for materials that exhibit essentially bilinear stress-strain behavior.

12.4 *Offset Shear Strength*—If desired, an offset shear strength may be determined from the shear stress versus shear strain curve. Translate the shear chord modulus of elasticity line along the strain axis from the origin by a fixed strain value and extend this line until it intersects the stress-strain curve. Determine the shear stress that corresponds to the intersection point and report this value, to three significant digits, as the offset shear strength, along with the value of the offset strain, as in:

$$F_{12}^{\circ} (0.2 \% \text{ offset}) = 70 \text{ MPa} \quad (6)$$

A graphical example of offset shear strength is shown in Fig. 3.

NOTE 5—In the absence of evidence suggesting the use of a more appropriate value, an offset strain value of 0.2 % is recommended.

12.5 *Statistics*—For each series of tests calculate the average value, standard deviation, and coefficient of variation (in percent) for each property determined:

$$\bar{\chi} = \left(\sum_{i=1}^n \chi_i \right) / n \quad (7)$$

$$s_{n-1} = \sqrt{\left(\sum_{i=1}^n \chi_i^2 - n\bar{\chi}^2 \right) / (n-1)} \quad (8)$$

$$CV = 100 \times s_{n-1} / \bar{\chi} \quad (9)$$

where:

- $\bar{\chi}$ = sample mean (average),
- s_{n-1} = sample standard deviation,
- CV = sample coefficient of variation, in percent,
- n = number of specimens, and
- χ_i = measured or derived property.

13. Report

13.1 The data reported with this test method include mechanical testing data, material identification data, fiber filler, and core material identification data and shall be in accordance with Guides E 1434, E 1309, and E 1471, respectively. Each

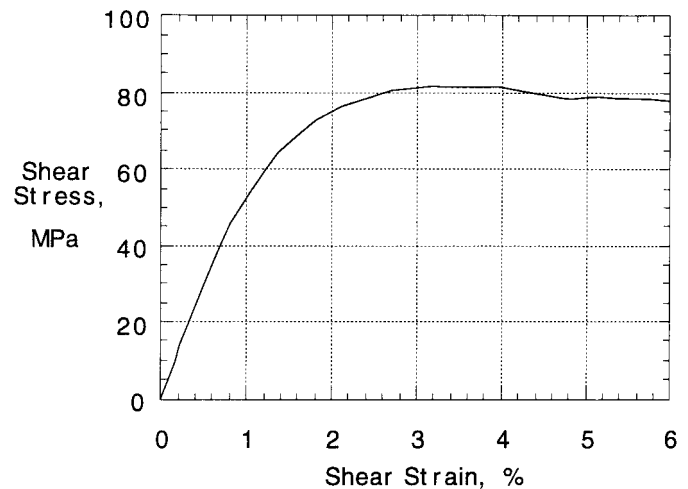


FIG. 4 Typical Shear Stress-Shear Strain Curve for PMC with Low-Ductility Matrix

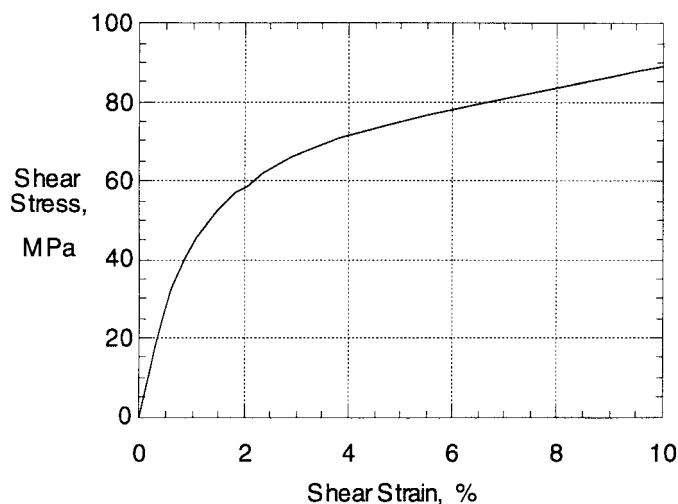


FIG. 5 Typical Shear Stress-Shear Strain Curve for PMC with Ductile Matrix

data item discussed is identified as belonging to one of the following categories: (VT) required for reporting of a valid test result, (VM) required for valid material traceability, (RT) recommended for maximum test method traceability, (RM) recommended for maximum material traceability, or (O) for optional data items. At a minimum, the report shall include all (VT) category items from Guide E 1434.

13.1.1 Clarification of Guide E 1434 Responses for This Test Method:

13.1.1.1 Field A1, Test Method—The response shall be either “D 3518 – 94” or “D 3518M – 94,” as appropriate.

13.1.1.2 Field A5, Type of Test—The response shall be “in-plane shear.”

13.1.1.3 Field B2, Specimen Orientation—The response shall be “0.0.”

13.1.1.4 Block E, Transducer Block—Used twice; once for each transducer.

13.1.1.5 Block F, Specimen Geometry Block—F6 (reinforcement volume) may be actual values, or it may be the average value for a sample. F7 (overall length) and F8 (gage length) may be actual values, or they may be the nominal or average value for the sample. F9 (area) is the actual area in accordance

with Test Method D 3039/D 3039M. F10 must also satisfy Test Method D 3039/D 3039M.

13.1.1.6 H32/K58, Progressive Damage Parameter—The response shall be “0.2 % offset strength.”

13.2 In addition to the data reported in accordance with Test Method D 3039/D 3039M, the report shall include the following information:

13.2.1 The revision level or date of issue of this test method.

13.2.2 Shear strain range used for chord shear modulus determination.

13.2.3 If another definition of shear modulus of elasticity is used in addition to chord shear modulus, describe the method used, the resulting correlation coefficient (if applicable), and the shear strain range used for the evaluation.

13.2.4 Individual values of shear modulus of elasticity, and the average, standard deviation, and coefficient of variation (in percent) values for the population.

13.2.5 Individual values of offset shear strength with the value of the offset strain, along with the average, standard deviation, and coefficient of variation (in percent) values for the population.

13.2.6 Individual maximum shear stresses, and the average, standard deviation, and coefficient of variation (in percent) values for the population. Note any test in which the failure load was less than the maximum load before failure.

13.2.7 Individual maximum shear strains, and the average value, standard deviation, and coefficient of variation (in percent) for the population. Note any test that was truncated to 5 % shear strain.

14. Precision and Bias

14.1 Precision—The data required for the development of a precision statement is not available for this test method. Committee D30 is currently planning a round-robin test series for this test method to determine precision.

14.2 Bias—Bias cannot be determined for this test method as no acceptable reference standard exists.

15. Keywords

15.1 composite materials; shear modulus; shear properties; shear strength

APPENDIX

(Nonmandatory Information)

X1. SIGNIFICANT POINTS OF MAJOR REVISIONS TO THIS TEST METHOD

X1.1 1991 Revision:

X1.1.1 Updated the format to conform to 1989 Form and Style for ASTM Standards.

X1.1.2 Changed the title and clarified the scope.

X1.1.3 Transformed the document from an inch-pound standard to a dual-units standard.

X1.1.4 Updated the Terminology section.

X1.1.5 Added a new Interferences section.

X1.1.6 Added a new Calibration section.

X1.2 1994 Revision:

X1.2.1 Updated the format to conform to current practices of Committee D30.

X1.2.2 Relaxed the scope to allow usage of this test method on material forms reinforced by woven fabrics.

X1.2.3 Extensively updated the Interferences section to

D 3518/D 3518M

discuss difficulty with this test method in obtaining reliable values for ultimate shear strength, and generally replaced the modifier ultimate with the term maximum throughout the text, as appropriate.

X1.2.4 Added to the coupon geometry, limitations on the stacking sequence of the test specimen laminate, and provided recommended values for coupon length and width.

X1.2.5 Indirectly incurred several procedural changes, through changes to Test Method D 3039/D 3039M, including the approach to standard conditioning described by Test Method D 5229/D 5229M.

X1.2.6 Added the determinations and reporting of an offset shear strength.

X1.2.7 Shear modulus reporting now includes, at a minimum, a strain-range-based chord shear modulus and also requires documentation of the strain range used for any other modulus definitions reported.

X1.2.8 Shear stress/shear strain data has been truncated at 5 % shear strain.

X1.2.9 Ultimate shear strength has been removed from this practice, replaced by reporting of maximum shear stress, determined as the maximum shear stress not exceeding 5 % shear strain.

X1.2.10 New illustrations have been included.

X1.2.11 Data reporting now follows Guide E 1434.

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